

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION
Ames Research Center
Moffett Field, California
PIONEER VENUS PROJECT
SPECIFICATION PC-456.04
PIONEER VENUS: DATA USER REQUIREMENTS
FOR SUPPLEMENTARY EXPERIMENTER DATA RECORDS
May 15, 1976

1. SCOPE
This specification defines the requirements of each of the Pioneer Venus data users for the Supplementary Experimenter Data Records (SEDR).
2. APPLICABLE DOCUMENTS
 - 2.1 NASA/ARC SPECIFICATIONS
 - PC-456.00, Pioneer Venus: Data Records Processing System (DRPS) Description
 - PC-456.01, Pioneer Venus: Input and Output Tape Description
 - PC-456.02, Pioneer Venus: Data Records Processing System (DRPS) Detailed Processing Requirements
 - PC-456.03, Pioneer Venus: Data User Requirements for Experimenter Data Records
 - PC-454.00, Pioneer Venus: On-Line Ground Data System Software Specification
 - PC-455.00, Pioneer Venus: Off-Line Telemetry Data Processing Specification

Section No. 3.0
Doc. No. YL-430.U4
Orig. Issue Date 7/20/77
Revision No. 3
Revision

3. REQUIREMENTS

3.1 GENERAL REQUIREMENTS

The general requirements and content for the Supplementary Experimenter Data Records have been covered in PC-456.01 Pioneer Venus: Input and Output Tape Requirements.

3.2 SPECIFIC REQUIREMENTS

The SEDR formats for all data users shall conform to the specification given below. In the case of the Orbiter Spacecraft, the SEDR shall be generated periodically from launch through mission completion. The Multiprobe Spacecraft SEDR shall be generated periodically from launch up until probe separation. A predicted model trajectory shall be generated for each of the probes after separation, from planet entry to impact in a format TBS.

Details on the use of the various tables defined in the SEDR format and the QLSEDR format are described in Section 6.3 of this specification. Coordinate System diagrams in the trajectory portion of the format are in Figures 3.2 Sheets 1 through 8.

3.2.1 Production SEDR. The SEDRs— for both spacecraft shall contain measured or true trajectory data as shown in Figure 3.2.1 sheets 1 through 10. These data shall be in IBM 360 Double Precision Floating Point formats.

3.2.2 Quick Look SEDR (QLSEDR). The best available data at the time the and will not contain the resolution that the production SEDRs will contain. The Logistics file shall be identical to that described above. The data file shall contain only 24 of the variables that are in the normal production SEDR. Since the variables are predicted, these selected variables shall be in IBM 360 Single Precision Floating Point formats. The format and variables which shall be carried in the QLSEDR are shown in Figure 3.2.2 sheets 1 through 7.

3.3 SEDR TAPE DELIVERY

SEDR tapes shall be sent to each addressee shown in Figure 3.4 of Specification PC-456.03.

Section No. 6.3.1.2.1
Doc. No. PC-456.04
Orig. Issue Date 7/20/77

Revision No. 3

Revision

6.3.1.2.1 Measured SRR Times. For some SRR pulses, one of the telemetered ATTM measurements will contain the time interval between the SRR pulse and the beginning of the major telemetry frame which immediately follows. For these SRR pulses, the time of the SRR pulse is calculated by:

$$TSRR = TMAJ - (\text{telemetered time interval})$$

TMAJ is the Spacecraft Universal Time (SCUT) of the beginning of the major telemetry frame.

6.3.1.2.2 Estimated SRR Times. For about half of the SRR pulses, no time interval measurement will be made. For these pulses, the time of the SRR pulse is estimated by:

$$TSSR(\text{est}) = TSRR (\text{last measured}) + (n) (\text{spin period})$$

TSRR (last measured) is the SCUT of the last SRR pulse for which a measured time interval was telemetered.

Spin period is the spacecraft spin period from Table 3. n is the integer number of spin periods which have elapsed since the last measured SRR pulse.

6.3.1.3 Smoothed Roll Reference Times (Fs). An Fs signal follows each SRR pulse. The time of the Fs signal is calculated by:

$$TFs = TSRR + \text{delta}$$

delta is the time delay from SRR to Fs from Table 3.

6.3.1.4 Roll Index Pulse Times (RIP). A RIP pulse follows each Fs signal.

The time of the RIP pulse is calculated by:

$$\text{TRIP} = \text{TFS} + (\text{RIP time delay})$$

(RIP time delay) is the current value of the time from Fs to RIP as determined from the telemetered measurement ARIPAD. (RIP time delay) = (decimal value of ARIPAD) (Spin Period)/1024

Spin period is the spacecraft spin period from Table 3.

Section No. 6.3.1.5
Doc. No. PC-456.04
Orig. Issue Date 7/20/77
Revision No. 3

6.3.1.5 Ram Pulse Times (RAM). A RAM pulse follows each RIP pulse. The time of the RAM pulse is calculated by:

$$\text{TRAM} = \text{TRIP} + (\text{RAM time delay})$$

(RAM time delay) is the value of the RAM time delay at the RIP time. The RAM time delay at the time of the beginning of each major telemetry frame is calculated from the telemetered measurement AVPDRD by:

$\text{RAM time delay} = (\text{decimal value of AVPDRD}) \times (\text{spin period}) / 4096$
The RAM time delay at the RIP time is interpolated to the RIP time from the telemetered measurements.

(spin period) is the spacecraft spin period from Table 3.

6.3.1.6 NADIR Pulse Times (NADIR). A NADIR pulse follows each RIP pulse. The time of the NADIR pulse is calculated by:

$$\text{TNADIR} = \text{TRIP} + (\text{NADIR time delay})$$

(NADIR time delay) is the value of the NADIR time delay at the RIP time.

The NADIR time delay at the time of the beginning of each major telemetry frame is calculated from the telemetered measurement AVPDND by:

$$\text{NADIR time delay} = (\text{decimal value at AVPDND}) \times (\text{spin period}) / 4096$$

(spin period if the spacecraft spin period from Table 3).

The NADIR time delay at the RIP time is interpolated to the RIP time from the telemetered measurements.

Section No. 6.3.2
Doc. No. PC-456.04
Orig. Issue Date 7/20/77
Revision No. 3
Revision

6.3.2 Pulse Times - Table 1. The times of one complete set of roll reference signals shall be included in Table 1 of the SEDR tape and QL Tape. For the Orbiter spacecraft, one complete set of roll reference signals shall include an Fs signal, the RIP pulse which is based on that Fs signal and the RAM and NADIR pulses which are based on the RIP pulse. For the Bus spacecraft, one complete set of roll reference signals shall include an Fs signal and the RIP pulse which is based on that Fs signal. A separate data record shall be included in the tape for each set of roll reference signals which occur during the time span of the data.

The signal times shall be the SCUT of each signal which have been calculated from the telemetry record as described in section 6.3.1.

6.3.3 Spacecraft Attitude. Table 2 shall tabulate the celestial latitude and celestial longitude of the spacecraft spin axis. The entries in the table will be provided as input and will not be calculated. Interpolation for spin axis attitude at times between the tabulated times shall be by using the technique described in section 6.3.6.

6.3.4 Spacecraft Spin Period - Table 3. Table 3 shall tabulate the spacecraft spin period and associated time delay (section 6.3.1.3). The entries in this table will be provided as input and will not be calculated. Linear interpolation may be used to estimate spin period and time delay at times between the entries in the table.

6.3.5 SRR Epochs - Table 4. Table 4 shall tabulate the positions of roll reference celestial bodies. The entries in this table will be input and will not be calculated. If the telemetry measurement ASRRMS indicates that a star is being used for the roll reference, the star celestial latitude and celestial longitude will be included in Table 4. If the telemetry measurement ASRRMS indicates that the Simulated Roll Reference (SIM SRR) is being used, the error

between the Sun and the Sim SRR will be tabulated. Table 4 will contain enough entries to model any Sim SRR drift and allow linear interpolation between entries.

The use of Table 4 to calculate spacecraft roll angle is described in section 6.4.

Section No. 6.3.6
Doc. No. PC-456 04
Orig. Issue Date 7/20/77
Revision No. 3

Revision

6.3.6 Vector Interpolation for Spacecraft Attitude.

6.3.6.1 Interpolation Requirement. If the spacecraft spin axis attitude is needed at a time which is between the times for which Table 2 contains tabulated values, the intermediate attitude must be interpolated as a vector: the attitude must be interpolated in the plane which contains the two tabulated vectors.

6.3.6.2 Vector Representation of Attitudes. Let T be the time at which the interpolated attitude is desired. Let T1 and T2 be the times at which the spin axis attitude is tabulated in Table 2.
[T1 < T < T2]

ATT1 = vector spin axis attitude at time T1

ATT2 = vector spin axis attitude at time T2

ATT = vector spin axis attitude at time T

ATT1X, ATT1Y, ATT1Z = X, Y, Z component of ATT1

ATT1X = cos (CLAT1) cos (CLON1)

ATT1Y = cos (CLAT1) sin (CLON1)

ATT1Z = sin (CLAT1)

ATT2X, ATT2Y, ATT2Z = X, Y, Z components of ATT2

ATT2X = cos (CLAT2) cos (CLON2)

$$\text{ATT2Y} = \cos (\text{CLAT2}) \sin (\text{CLONZ})$$

$$\text{ATT2Z} = \sin (\text{CLAT2})$$

$\text{ATTX}, \text{ATTY}, \text{ATTZ}$ = X, Y, Z components of the interpolated vector ATT

Figure 6.3.6.2 shows the vector relations.

Section No. 6.3.6.3
 Doc. No. PC-456.04
 Orig. Issue Date 7/20/17
 Revision No. 3
 Revision

6.3.6.3 Minimum Angle between Vectors. Check that the angle between ATT1 and ATT2 is large enough to avoid a zero divisor later.

$$\text{MAGDIF} = [(ATT2X - ATT1X)^2 + (ATT2Y - ATT1Y)^2 + (ATT2Z - ATT1Z)^2]^{1/2}$$

If MAGDIF less than 0.00017 set

ATTX = ATT1X
 ATTY = ATT1Y
 ATTZ = ATT1Z

6.3.6.4 Interpolation Equations.

BIGANG = 2 sin⁻¹ (MAGDIF/2)
 RATIO = (T-T1)/(T2 - T1)
 LILANG = (BIGANG) (RATIO)
 ATTX = (ATT2X - ATT1X cos (BIGANG)) RATIO + ATT1X cos (LILANG)
 ATTY = (ATT2Y - ATT1Y cos (BIGANG)) RATIO + ATT1Y cos (LILANG)
 ATTZ = (ATT2Z - ATT1Z cos (BIGANG)) RATIO + ATT1Z cos (LILANG)

6.3.6.5 Interpolated Attitude.

CLAT = interpolated celestial latitude
 CLAT = sin⁻¹ (ATTZ)

CLON = interpolated celestial longitude

CLON = tan⁻¹ (ATTY)/ATTX

6.3.6.6 Attitude Data. When spacecraft attitude is interpolated to the epochs of the SEDR Ephemeris, include the components of ATT in the Ephemeris Data File.

Interpolated component	Ephemeris File Variable	Ephemeris File Words
ATTX	63	124 and 125
ATTY	64	126 and 127
ATTZ	65	128 and 129

When spacecraft attitude is interpolated to the epochs of the Quick Look Data Record, include the components of ATT in the Ephemeris Record, File 6.

Section No. 6.3.6.6
Doc. No. PC-454.04
Orig. Issue Date 7-20-77
Revision No. 3
Revision

Interpolated component	Ephemeris File Word
ATTX	18
ATTY	19
ATTZ	20

	PRL	LRL	P/L	FILE I.D.
FILE 1				LOGISTICS RECORD (EBCDIC)
				EOF
FILE 2				PT11 RECORD
				EOF
FILE 3				AT12 RECORD
				AT12 RECORD
				EOF
FILE 4				SP13 RECORD
				5P13 RECORD
				EOF

PRL	LRL	P/L	FILE I.D.
			SR14 RECORD
			SR14 RECORD
			EOF
PRL	LRL	P/L	FILE I.D.
			EPHEMERIS RECORD (ORBIT TODAY)
FILE 6			EPHEMERIS RECORD
			EOF
			EOF

PIONEER VENUS SEDR TAPE
 FORMAT DEFINITION
 REV.NO. 3 DATE 1/20/

For Definitions
 Refer to Log
 File Table

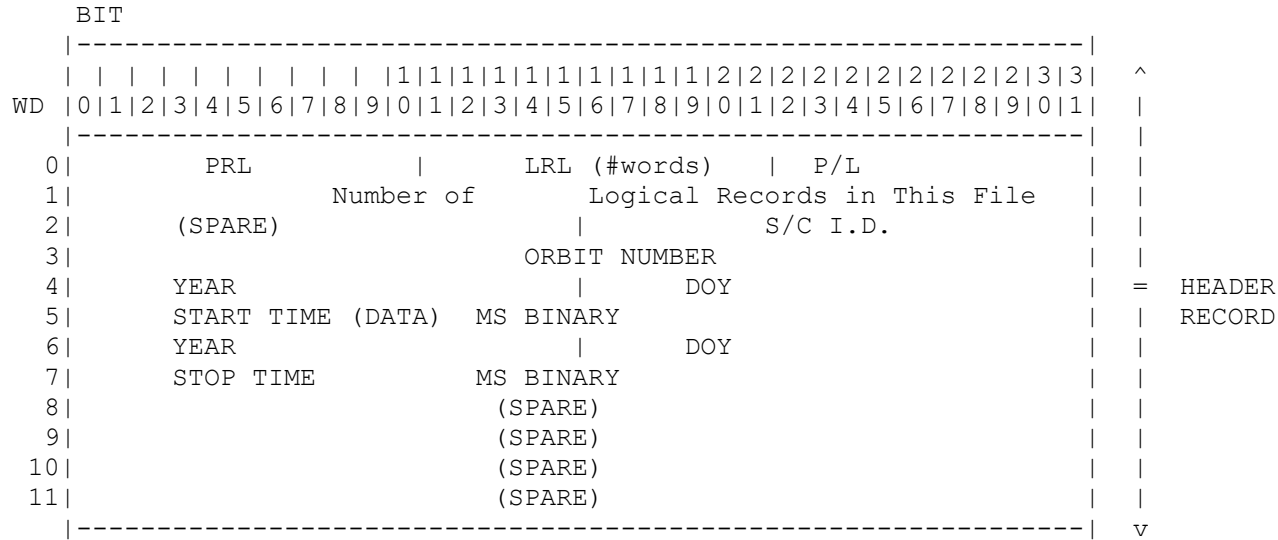
	WORD 0	WORD 1	WORD 2	WORD 3	WORD 4	WORD 5	
RECORD 1	*LH WD						
2	* P I O N E E R	V E N U S	S E D R *				
3							
4	T A P E	S E Q U E N C E	N O . X X X X				7
5							
6							1
7	S / C I D					X X	7
8	O R B I T	N O				X X X X	7
9							
10							
11							
12							
13							
14							
15							
16	G E N E R A T E D		M M M		D D , Y Y		2,7
17	S T A R T	D A T E			D O Y , Y Y		7
18	S T A R T	T I M E			H H : M M		7
19	S T O P	D A T E			D O Y , Y Y		7
20	S T O P	T I M E			H H : M M		7
21							
22	D Q I	P (O R)	A				8

WORD 0 OF RECORD 1 IS LOGICAL READER WORD

*LH WD = LOGICAL HEADER WORD
 DATA FORMAT: EBCDIC
 (LRL) LOGICAL RECORD LENGTH: 6 WORDS
 *LOGICAL RECORDS IN FILE: 22
 (PRL) PHYSICAL RECORD LENGTH: 132
 LOGICAL RECORDS PER PHYSICAL RECORD: 22

PIONEER VENUS SEDR FILE DEFINITION
 LOGISTIC FILE RECORD FORMAT
 REV.NO. 3 DATE 1/20/78

Fig 3.2.1



BIT																																			

	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1			
WD	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1			

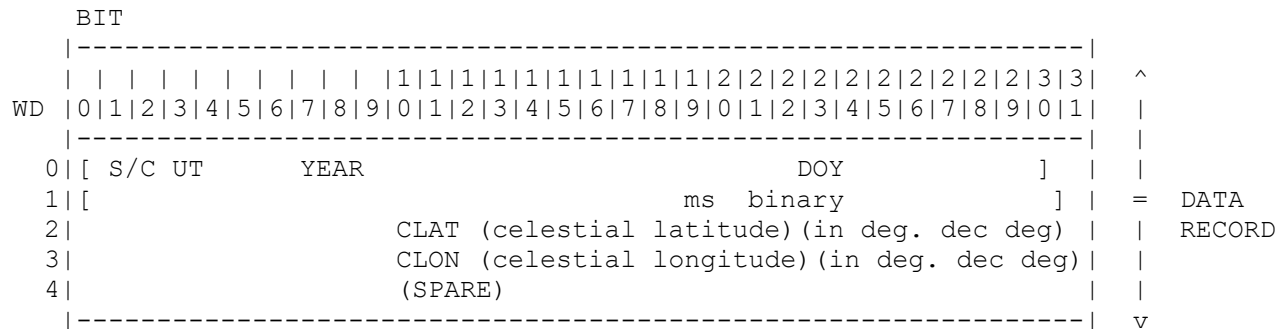
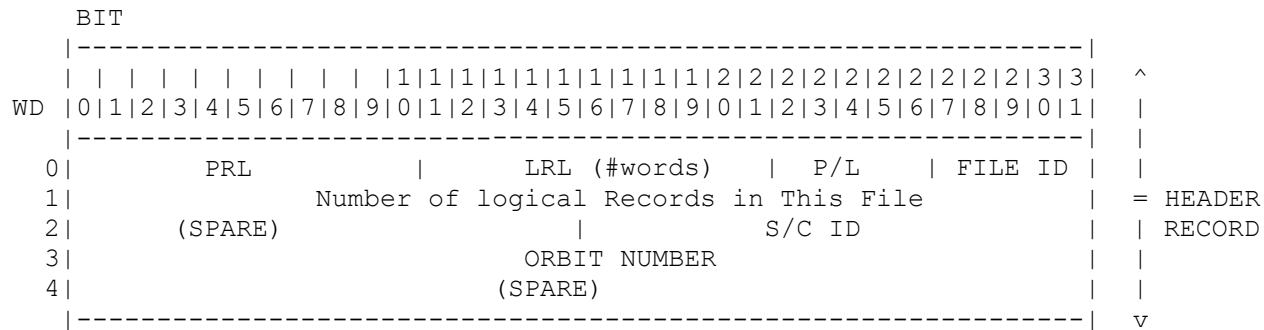
0	[YEAR											DOY]												
1	[ms binary										(S/C UT of Fs Pulse]																						
2	[YEAR											DOY]												
3	[ms binary										(S/C UT of RIP Pulse)]																						
4	[YEAR											DOY]		=	DATA									
5	[ms binary										(S/C UT of RAM Pulse]			RECORD																			
6	[YEAR											DOY]												
7	[ms binary										(S/C UT of NADIR Pulse]																						
8		DATA QUALITY FLAG 1 Delta - R*8																																	
9		(SPARE)																																	
10		A													B		C		D																
11		UCLA Pulse time correction (EBCDIC)																																	
	-----																																	v	

Logical Record Length : 12 words A: STRM
 Data format : Binary B: CICK
 Physical Record Length : 120 words C: SRRM
 No. Logical Records/Physical Record : 10 D: Fire

SRRM: 0 = unk Fire: 0 = Norm
 1 = Star (22,23) 1 = Fire
 2 = Sim
 3 = Sun

PIONEER VENUS SEDR TAPE FILE
 DEFINITION FILE 2
 TABLE 1 - PULSE TIMES (P111)
 LOGICAL RECORD FORMAT
 REV. NO. 3 DATE 1/20/78

FIG 3.2.1



Logical Record Length : 5 words
 Data Format : Single Precision Flt. Pt. Binary
 Physical Record Length: 50 words
 Logical Records/Physical Record : 10

PIONEER VENUS SEDR TAPE FILE
 DEFINITION FILE 3
 TABLE 2 - S/C ATTITUDE (AT12)
 REV. NO. 3 DATE 1/20/78

FIG 3.2.1

BIT				

	1 1 1 1 1 1 1 1 1 1 2 2 2 2 2 2 2 2 2 2 3 3			^
WD	0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1			

0	PRL		LRL (# words)	P/L FILE I.D
1			Number of	Logical Records in This File
2	(SPARE)			S/C I.D.
3			ORBIT NUMBER	= HEADER
4	YEAR			DOY
5	START DATA TIME			MS BINARY
6	YEAR			DOY
7	STOP DATA TIME			MS BINARY
8			SPARE	
9			SPARE	
	-----			v

PIONEER VENUS SEDR TAPE FILE
 DEFINITION FILE 4
 TABLE 3 - S/C SPIN RECORD (SP13)
 LOGICAL RECORD FORMAT
 REV. NO. 3 DATE 1/20/78

BIT																																	

	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	
WD	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	
0	PRL			LRL (# words)			P/L			FILE I.D																							
1	Number of						Logical Records in This File																										
2	(SPARE)						S/C I.D.																										
3	ORBIT NUMBER						= HEADER																										
4	YEAR						DOY																										
5	START DATA TIME						MS BINARY																										
6	YEAR						DOY																										
7	STOP DATA TIME						MS BINARY																										
8	SPARE																																
9	SPARE																																
-----														v																			

PIONEER VENUS SEDR TAPE FILE
 DEFINITIONS FILE 5
 TABLE 4 - SRR EPOCHS (SR14)
 LOGICAL RECORD FORMAT
 REV. NO. 3 DATE 1/20/78

EPHEMERIS HEADER RECORD

BIT 1 1 1 1 1 1 1 1 1 1 2 2 2 2 2 2 2 2 2 2 3 3
 0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1

word 0	PRL	LRL (# words)	P/L	FILE ID
1	Number of Logical Records in This File			
2	(spare)		S/C I.D.	
3	ORBIT NUMBER			
4	S/C UT YEAR		DOY	S/C UT
5	START TIME - MILLISECONDS			
6	S/C UT YEAR		DOY	S/C UT
7	STOP TIME - MILLISECONDS			
8	SPARE			
9	SPARE			
	.			
	.			
	.			
282	SPARE			
283	SPARE			

All data is in IBM 360 binary format, 2 and 4 byte integers

EPHEMERIS DATA RECORD

LOGICAL RECORD LENGTH: 284
 PHYSICAL RECORD LENGTH: 284

1. ALL DATA IS IN IBM 360 format, DP
 2. Except for integer variables, all variables are in double-precision floating point (data records). including calculated attitude, nadir and ram.

NO. OF LOGICAL RECORDS PER PHYSICAL RECORD: 1

3. File order is in S/C UT ascending order.

BIT 1 1 1 1 1 1 1 1 1 1 2 2 2 2 2 2 2 2 2 3 3
 0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1

WORD

WORD	0	1	2-3	4-5	6-7	8-9	10-11	12-13	14-15	16-17	18-19	20-21
	S/C	YEAR			S/C DOY		NAME	VAR				
	S/C	UT in binary					SCUT	1				
	DP Julian Date, Days (elapsed days from Noon, 1 Jan. 4713 B.C.)							JULDAT	2			
	Gregorian Calendar Date (2 integer words which represent the vigesimal date) based on number 20							VIGDAT	3			
	ET - UTC, set (Universal Time Co-ordinated)							ETMUTC	4			
	Range rate of probe. km/sec						+	RANGRF	5			
	Magnitude of velocity vector, km/sec (speed of S/C)						Geocentric	MAGVEL	6			
	Earth.probe range, km (radius to S/C)						+	REARPR	7			
	Range to probe, km						+	MRANGE	8			
	Probe inertia velocity, km/sec							MMAGVF	9			
	Probe inertial path angle, deg						HELIO-CENTRIC	HINFTP	10			
	Celestial latitude of probe, deg							CELLTF	11			

22-23	Celestial longitude of probe, deg		CELLNF 12
24-25	Celestial latitude of Earth, deg		CELLTE 13
26-27	Celestial longitude of Earth, deg		CELLNE 14
28-29	X component of S/C in Sun-Earth line, km	HELIO-	XSCSEL 15
30-31	Y component of S/C in Sun.Earth line, km	CENTRIC	YSCSEL 16
32-33	Z component of S/C in Sun.Earth line, km		ZSCSEL 17
34-35	Sun-probe distance in Sun-Earth X-Y plane, km (projection of Sun-S/C vector onto the X-Y plane		SPSE 18
36-37	Longitude of S/C in Sun-Earth line system, deg	+	LNPSEL 19

Note: The term "probe" refers to the Multiprobe or Orbiter spacecraft and not to the separated Large or Small Probes.

Fig 3.2.1

EPHEMERIS DATA RECORD con't

WORD	BIT	1	1	1	1	1	1	1	1	1	2	2	2	2	2	2	2	2	3	3	name	var	
	0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	
38-39		X	component	of	geocentric	r	of	S/C,	km		+											XPGSFF 20	
40-41		Y	component	of	geocentric	r	of	S/C,	km														YPGSFF 21
42-43		Z	component	of	geocentric	r	of	S/C,	km														ZPGSFF 22
44-45		X	component	of	geocentric	r	of	S/C,	km/sec														DXPGSF 23
46-47		Y	component	of	geocentric	r	of	S/C,	km/sec														DYPGSF 24
46-49		Z	component	of	geocentric	r	of	S/C,	km/sec														DZPGSF 25
50-51		X	component	of	heliocentric	r	of	S/C,	km														XPHSFF 26
52-53		Y	component	of	heliocentric	r	of	S/C,	km			Space-fixed											YPHSFF 27
54-55		Z	component	of	heliocentric	r	of	S/C,	km			Coordinates											ZPHSFF 28
56-57		X	component	of	heliocentric	r	of	S/C,	km/sec			FERP											DXPHSF 29
56-55		Y	component	of	heliocentric	r	of	S/C,	km/sec			Earth Mean											DYPHSF 30
60-61		Z	component	of	heliocentric	r	of	S/C,	km/sec			Equinox &											DZPHSF 31
62-63		X	component	of	body 1 - S/C	r,	km				+												XP1SFF 32
64-65		Y	component	of	body 1 - S/C	r,	km																YP1SFF 33
66-67		Z	component	of	body 1 - S/C	r,	km																ZP1SFF 34
68-69		X	component	of	body 1 - S/C	r,	km/sec					Body											DXP1SF 35
70-71		Y	component	of	body 1 - S/C	r,	km/sec					1 = Venus											DYP1SF 36
72-73		Z	component	of	body 1 - S/C	r,	km/sec																DZP1SF 37

74-75	Body 1 - S/C range, km			B1MAGR	38
76-77	Body 1 - S/C velocity magnitude, km/sec	++		B1MAGV	39
78-79	Latitude of probe, deg	+		EALATP	40
80-81	Longitude of probe, deg			EALOMP	41
82-83	Velocity magnitude of probe relative to body km/sec.	(see EN/1)	Body Fixed Earth	EAVELP	42
84-85	Body fixed path angle of probe. deg			EAPTHP	43
86-87	Body fixed azimuth angle of probe. deg	+		EAAZIP	44

Note: The term "probe" refers to the Multiprobe or Orbiter spacecraft and not to the separated Large or Small Probes.

EN/1 -- Earth true equator of date and Greenwich Meridian

Fig 3.2.1

EPHEMERIS DATA RECORD con't

BIT		1 1 1 1 1 1 1 1 1 1 2 2 2 2 2 2 2 2 2 2 3 3																								
WORD		0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1																				name	var			
88-89	Latitude of probe. deg																			+			B1LATP	45		
90-91	Longitude of probe. deg																						B1LOMP	46		
92-93	Velocity magnitude of probe relative to body km/sec																				Body Fixed			B1VELP	47	
																					Body 1 = Venus					
																					(see EN/2)					
94-95	Body fixed path angle of probe. deg																						B1PTHP	48		
96-97	Body fixed azimuth angle of probe. deg																			+			B1AZIP	49		
100-101	Earth-probe-sun angle, deg																						EPSUAN	51		
102-103	Sun-Earth-probe angle, deg																				Angle Group			SEPANG	52	
																					All angles in					
																					degrees					
104-105	Earth-Sun-probe angle, deg																						ESPANG	53		
106-107	Sun-probe-body 1 angle, deg																						SPB1AN	54		
108-109	Body 1-Earth-probe angle, deg																			+			B1EPAN	55		
110-111	Flag for periapsis (See EN/3)																						PERIAP	56		
112-113	X-coordinate of XROLL																			+	+			XROLLX	57	
114-115	Y-coordinate of XROLL																						XROLLY	58		
116-117	Z-coordinate of XROLL																						XROLLZ	59		
118-119	X-coordinate of YROLL																						YROLLX	60		
120-121	Y-coordinate of YROLL																				Attitude	Spacecraft -			YROLLY	61
																					Data	Centered				
																						Non-Rotating				
122-123	Z-coordinate of YROLL																						YROLLZ	62		
																					Coordinates					

124-125	X-coordinate of ATT			ATTX	63
126-127	Y-coordinate of ATT			ATTY	64
128-129	Z-coordinate of ATT	+		ATTZ	65
130-131	NADROL			NADROL	66
132-133	NADLOK			NADLOK	67
134-135	RAMROL			RAMROL	68
136-137	RAMLOK		+	RAMLOK	69

Note: The term "probe" refers to the Multiprobe or Orbiter spacecraft and not to the separated Large or Small Probes.

EN/2 -- Venus true equator of date and Prime Meridian
 EN/3 -- PERIAP = 0. for no closest approach,
 = 1 for periapsis to PCB,
 = 2 for apoapsis to PCB

Fig 3.2.1

EPHEMERIS DATA RECORD con't

BIT		1 1 1 1 1 1 1 1 1 1 2 2 2 2 2 2 2 2 2 2 3 3																				name	var
WORD		0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1																					
138-153	SPARES																					70-77	
154-155	Celestial latitude of probe, deg										+		DECPI	78									
156-157	Celestial longitude of probe, deg												RAP1	79									
158-159	Flight-path angle of probe, deg												PTHP1	80									
160-161	Azimuth angle of probe, deg												AZP1	81									
162-163	Range rate of probe. km/sec												DR1	82									
164-165	X-component of Earth position, km											(EC)	XE1	83									
166-167	Y-component of Earth position, km											(EC)	YE1	84									
168-169	Z-component of Earth position, km											(EC)	ZE1	85									
170-171	X-component of Earth velocity, km/sec											(EC)	DXE1	86									
172-173	Y-component of Earth velocity, km/sec											(EC)	DYE1	87									
174-175	Z-component of Earth velocity, km/sec											(EC)	DZE1	88									
176-177	Venus-to-Earth range, km											Venus Centered Earth Mean	RE1	89									
178-179	Celestial latitude of Earth direction, deg											Equinox & Ecliptic of	DECE1	90									
180-181	Celestial longitude of Earth direction, deg											1950.0	RAE1	91									
182-183	X-component of Sun position. km											(EC)	XS1	92									
184-185	Y-component of Sun position, km											(EC)	YS1	93									
186-187	Z-component of Sun position, km											(EC)	ZS1	94									
188-189	X-component of Sun velocity, km/sec												DXS1	95									

190-191	Y-component of Sun velocity, km/sec	DYS1	96
192-193	Z-component of Sun velocity, km./sec	DZS1	97
194-195	Venus-to-Sun range, km	RS1	98
196-197	Celestial latitude of Sun direction, deg	DESC1	99
198-199	Celestial longitude of Sun direction. deg +	RAS1	100

Note: The term "probe" refers to the Multiprobe or Orbiter spacecraft and not to the separated Large or Small Probes.

Fig 3.2.1

EPHEMERIS DATA RECORD con't

BIT		1 1 1 1 1 1 1 1 1 1 2 2 2 2 2 2 2 2 2 2 3 3																												
WORD		0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1 2 3 4 5 6 7 8 9 0 1		name var																										
200-201	Orbit semimajor axis, km																											SMA	101	
202-203	Orbit eccentricity																											ECC	102	
204-205	Time from periapsis, secs																											TFP	103	
206-207	Radius of closest approach, km																											RCA	104	
208-209	orbit period, day																											PER	105	
210-211	Spacecraft true anomaly, deg																											TA	106	
212-213	Orbit inclination, deg																											INCL	107	
214-215	Longitude of ascending node, deg																											LANL	108	
216-217	Argument of periapsis, deg																											APF1	109	
218-219	X-component of unit vector to periapsis																											PX1	110	
220-221	Y-component of unit vector to periapsis																											PY1	111	
222-223	Z-component of unit vector to periapsis																											PZ1	112	
224-225	X-component of unit normal to orbit plane																											WX1	113	
228-227	Y.component of unit normal to orbit plane																											WY1	114	
228-229	Z-component of unit normal to orbit plane+																											WZ1	115	
230-231	Sun-Venus-probe angle, deg																												S200P	116
232-233	Sun-Earth-Venus angle, deg																												SE200	117
234-235	X-component of probe position, km																												XP1	118
236-237	Y-component of probe position, km																												YP1	119

238-239	Z-component of probe position, km		ZP1	120
240-241	X-component of probe velocity, km/sec		DXP1	121
242-243	Y-component of probe velocity, km/sec		DYP1	122
244-245	Z-component of probe velocity, km/sec	+	DZP1	123

Note: The term "probe" refers to the Multiprobe or Orbiter spacecraft and not to the separated Large or Small Probes.

Fig 3.2.1

EPHEMERIS DATA RECORD con't

BIT		1 1 1 1 1 1 1 1 1 1 2 2 2 2 2 2 2 2 2 3 3																										
WORD		0	1	2	3	4	5	6	7	8	9	0	1	2	3	4	5	6	7	8	9	0	1	name	var			
246-247	X-component of Earth position, km																							(EQ)	+		XE1	124
248-249	Y-component of Earth position, km																							(EQ)			YE1	125
250-251	Z-component of Earth position, km																							(EQ)			ZE1	126
252-253	X-component of Sun position, km																							(EQ)			XS1	127
254-255	Y-component of Sun position, km																							(EQ)		Venus Centered	YS1	128
256-257	Z-component of Sun position, km																							(EQ)		Earth Mean Equinox &	ZS1	129
258-259	X-component of probe position, km																									Ecliptic of 1950.0	XP2	130
260-261	Y-component of probe position, km																										YP2	131
262-263	Z-component of probe position, km																										ZP2	132
264-265	X-component of probe velocity, km/sec																										DXP2	133
266-267	Y-component of probe velocity, km/sec																										DYP2	134
268-269	Z-component of probe velocity, km/sec+																								+		DZP2	135
270-271	Celestial latitude of Earth-to-probe direction, deg (see EN/4)																										DECP3	136
272-273	Celestial longitude of Earth-to-probe direction, deg																										RAP3	137
274-279	(SPARES - Double Precision Zeros)																											138-140

EN/4 Earth-centered, Earth mean Equinox & Ecliptic of 1950.0

Fig 3.2.1

The next nine pages contain graphics. Please see TIFF files.